United States of America

Department of Transportation—Jederal Aviation Administration

Supplemental Type Certificate

Number SA4607NM

This certificate, issued to F. Atlee Dodge

certifies that the change in the type design for the following product with the limitations therefor as specified hereon meets the airworthiness requirements of Part 3 of the

Regulations.

Original Product - Type Certificate Number: A-780

Make: Piper

Model: PA-12 and PA-12S

Description of Trype Design Change:

Installation of main landing gear safety cable kit manufactured in accordance with the drawings listed on F. Atlee Dodge Drawing List 2089A, dated April 12, 1989, or later FAA approved revision, and installed per F. Atlee Dodge Installation Instructions Drawing 2089-12, dated 3-26-89, or later FAA approved revision.

Limitations and Conditions:

The seaplane versions of the Piper PA-12 airplanes are eligible for the safety cable kit installation when operating as landplanes with the standard PA-18 landing gear installed per an applicable STC.

(See Continuation Sheet Page 3 of 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a terminativn date is otherwise established by the Administrator of the Tederal Ariation Administration.

Date of application: 24 October 1972

Dute reissued :

Date of issuance: 19 April 1989

Date amended:

By direction of the Administrator

Alan C. Strickfader Signature) Supervisor Anchorage Aircraft Certification Field Office Northwest Mountain Region

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

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Supplemental Type Certificate

(Continuation Sheet)

Number SA4607NM

Limitations and Conditions: (Continued)

The approval of this change in type design applies basically piper Model PA-12 and PA-12S airplanes only. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will introduce no adverse effect upon the airworthiness of the aircraft. This determination should include consideration of significant change in weight distribution such as an increase in the fixed disposable weight in the fuselage.